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MARITIME PATROL AIRCRAFT (MPA) CONCEPT FORMULATION. ALLISON PD3--ETC(U)
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DDA-EDR-9774A

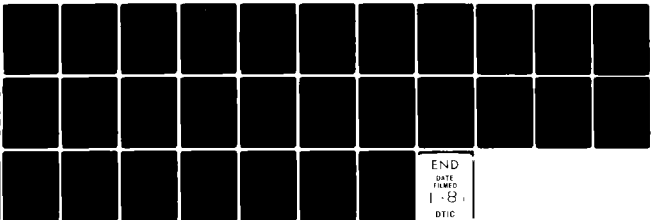
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| 20. ABSTRACT (Continue on reverse side if necessary and identify by block number) This study developed data on Detroit Diesel Allison (DDA) common core derivative engines for use in Maritime Patrol Aircraft (MPA) concept formulation studies. The study included the screening of potential DDA turboprop/turboshaft engines and the preparation of technical and planning information on three of the most promising engine candidates plus an all new engine. Screening of DDA derivative candidates was performed utilizing an analytical MPA model using synthesized mission profiles to rank the candidates in terms | | |

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of fuel consumption, weight, cost and complexity. The three turboprop engines selected for further study were as follows: a derivative of the unity size T701-AD-700 shaft power engine with rematched turbine (PD 370-37), an advanced T701 turboprop derivative with 25:1 overall pressure ratio and a scaled ATEC demonstrated compressor (PD 370-40), an advanced T701 turboprop derivative with 17.7:1 overall pressure ratio and a scaled ATEGG demonstrated compressor (PD 370-41). Data is also presented on a new advanced turboprop engine with 30:1 overall pressure ratio which incorporates compressor, combustor, turbine, and cooling technology now under development and demonstration at DDA. The documentation consists of six separate reports prepared in the following manner. One report summarizes the engine screening analysis and describes the approach to, and the conclusions of the study. A separate report for each of the three derivative engines and for the new turboprop present estimates of performance, weight, and dimensional data. The engineering budgetary estimates of the development, acquisition, and service costs for each of the four engines are presented in a separate report.

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REVISIONS

| <u>Letter</u> | <u>Page</u> | <u>Revision</u> |
|---------------|-------------|---|
| A | 4 | Gearbox and total weight |
| A | 5 | Gearbox, interconnecting struts and shaft, and total weight |
| A | 9 | Additional matrix points at 0 and 25,000 feet |
| A | 12 | Additional performance |
| A | 17 | Additional performance |
| A | 22 | Additional performance |
| A | 27 | Additional performance |



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I. INTRODUCTION

This report presents estimates of performance, weight, and dimensional data for the PD370-37 turboprop engine. The engine is in the 8000 to 9000 SHP class and is intended for subsonic aircraft. The data is submitted for use in preliminary design type studies in the evaluation of turboprop systems.

The PD370-37 is a derivative of the unity size T701-AD-700 shaft power engine, with the turbine rematched for better turboprop characteristics. The T701-AD-700 is a free turbine turboshaft engine that was developed through safety demonstration testing, for the U.S. Army's HLH program. The Model 570, a commercial industrial version of this engine, has undergone additional development testing, and is now in production.

The reduction gearbox for speed reduction to the prop-fan is a new simplified design compared to the DDA T56 series of gearboxes. The new design is based upon a study into the reliability and maintenance cost history of past turboprop systems, and follows the recommendations from that study for a gearbox with high reliability, easy maintainability, and low maintenance cost.



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II. ENGINE DESCRIPTION

The Model PD 370-37 is a single spool, free power turbine, axial flow power unit connected by shafting, and supporting structure to an offset reduction gear assembly. The general arrangement and external features of the engine are shown in Figure II-1, with principle physical characteristics listed in Table II-I. The reduction gear shown in Figure II-1 has an overall gear ratio of 10.83:1, providing a propfan speed of 1390 rpm at a constant power turbine output speed of 15,049 rpm. However, parametric weight data is shown in Section III so that other propfan rotational speeds, and gear ratios can be analyzed. An aircraft accessory drive pad is provided on the back of the gearbox to drive an aircraft mounted accessory drive box. Power available at this pad is 500 HP at 8000 rpm. The primary engine mounts are on the gearbox with a hang mount at the rear of the engine. Engine accessories are driven by a bevel drive from the high pressure spool. The control system is integral with the prop-fan and is electro-mechanical. The oil system is integral to the engine and also supplies the prop-fan and reduction gearbox, but is separately filtered and monitored to isolate fault detection in each of these major modules. Engine torque is measured hydraulically from the gear thrust of the power train idler gears in the reduction gearbox.

The gearbox is shown offset, based upon DDA's experience with large turboprop engines. It is offset-up to be consistent with current studies showing a preference to under-the-wing engine mounting. It can also be supplied in the offset-down position.

Performance ratings, sea level static, are listed in Table II-II.



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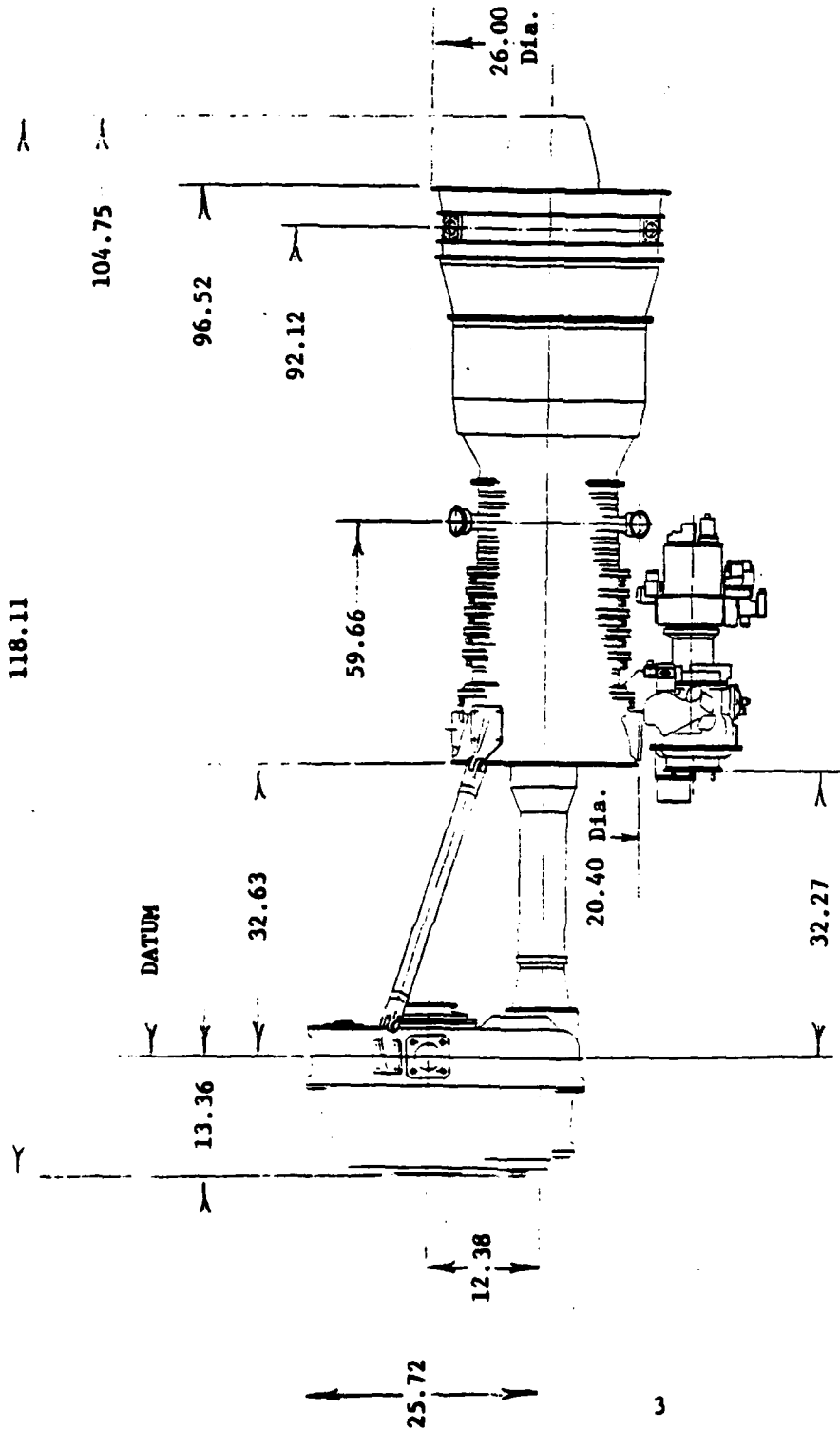


Figure II-1. PD370-37 General Arrangement.



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TABLE II-I

PD 370-37 PHYSICAL CHARACTERISTICS

(Includes Gearbox)

| | |
|---|--------|
| Length (in) | 118.11 |
| Max Engine Diameter (in) | 26.00 |
| Max Gearbox Offset, Upward (in) | 25.72 |
| Dry Weight, lbs. | |
| Engine | 1105 |
| Gearbox, including Interconnecting Struts & Shaft | 501 |
| Total | 1606 |

For gear ratios other than 10.83:1 the reduction gearbox dimensions may be scaled as follows:

$$\text{Dim}_{\text{GR}} = \text{Base dim} \times \left(\frac{\text{GR}}{10.83} \right)^{0.33}$$

Sea level performance ratings are summarized in Table II-II.

TABLE II-II

PD 370-37 PERFORMANCE SUMMARY

Sea Level, 0 Kts.

| | <u>Standard Day</u> | | | <u>Hot Day, 89.8°F</u> | | |
|-----------------|---------------------|-------|----------------|------------------------|-------|----------------|
| | SHP | SFC | F _N | SHP | SFC | F _N |
| Take-Off | 8450 | 0.443 | 1213 | 7766 | 0.448 | 1100 |
| Max. Continuous | 7455 | 0.443 | 1079 | 6408 | 0.461 | 913 |



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III. WEIGHTS

The weight of the basic engine, gearbox, interconnecting struts and shaft are given in Table III-I. The gearbox weight is based upon a gear ratio of 10.83:1 which provides a propfan speed of 1390 rpm.

TABLE III-I

PD 370-37 Weights

| | <u>Dry</u> | <u>Wet</u> [*] | <u>Installed</u> |
|----------------------------------|------------|-------------------------|------------------|
| Basic Engine, lbs. | 1105 | 1128 | 1128 |
| Gearbox, lbs. | 471 | 508 | 508 |
| Interconnecting Struts and Shaft | <u>30</u> | <u>30</u> | <u>30</u> |
| Total, lbs. | 1606 | 1666 | 1666 |

* Includes total amount of oil required for engine and gearbox operation.

For gear ratios other than 10.83:1, the gearbox, interconnecting strut and shaft dry weights may be estimated as follows:

$$\text{Dry Gearbox weight} = 471 \left(\frac{\text{GR}}{10.83} \right)^{0.4}$$

Interconnecting strut and shaft weight = 6.4% of dry gearbox weight.



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IV. STEADY STATE PERFORMANCE

Steady state performance data is tabulated in this section for all points shown in Figure IV-1. Basic engine data is shown for the following assumptions:

- o Uninstalled engine
- o ICAO standard atmosphere except for takeoff which in addition includes an ambient temperature of 89.8°F at standard atmosphere
- o 100% inlet recovery
- o Zero accessory horsepower extraction
- o Zero customer bleed extraction
- o Zero losses due to reduction gear
- o Fuel heating value - 18,400 Btu/lb
- o Estimated average engine performance - No SHP or fuel flow guarantee factors

Sensitivity data is provided for each point so that bleed and duct losses may be estimated as required.

Nomenclature

Nomenclature used in the tabulation of performance is as follows:

| | |
|------|---|
| MACH | Mach number |
| SHP | Shaft horsepower |
| SFC | Specific fuel consumption, lbs/hr/hp |
| WF | Engine fuel flow, lbs/hr |
| FN | Net jet thrust, lbs (jet gross thrust - ram drag) |
| ESHP | Equivalent shaft horsepower (energy in jet stream converted ideally to horsepower and added to SHP) |
| WCIN | Total inlet corrected airflow, $W\sqrt{\theta_1}/\delta_1$ |

where: θ_1 = Engine inlet total temp, °R
 $\frac{518.688}{\theta_1}$



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$$\delta_1 = \frac{\text{Engine inlet total pressure, psi}}{14.696}$$

| | |
|------|----------------------------------|
| TNOZ | Jet nozzle total temperature, °R |
| PNOZ | Jet nozzle total pressure, psi |
| RC | Compressor pressure ratio |
| BOT | Burner outlet temperature, °R |
| NO | Point number |

Sensitivity Data

Bleed:

SHP, with bleed = SHP, no bleed - (DEL SHP)(% bleed)

WF, with bleed = WF, no bleed - (DEL WF)(% bleed)

FN, with bleed = FN, no bleed - (DEL FN)(% bleed)

Inlet Recovery:

η_R = Total pressure actual/Total pressure ideal

SHP, with recovery = SHP, ideal recovery - (DEL SHP)(1 - η_R)(100)

WF, with recovery = WF, ideal recovery (η_R)

FN, with recovery = FN, ideal recovery - (DEL FN)(1 - η_R)(100)

Jet Nozzle Duct Loss:

To estimate thrust loss due to additional duct loss prior to the jet nozzle, use the following equation:

$$\text{FN, with loss} = \text{FN, without loss} - \text{FN, without loss} (K) \left(\frac{\Delta P}{P} \right)$$

where,

o K is obtained for each point from sensitivity data

$$o \frac{\Delta P}{P} = \frac{\text{PTOT, no loss} - \text{PTOT, total loss}}{\text{PTOT, no loss}}$$



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Reduction Gear Loss:

Reduction gear is 99 percent efficient.

Accessory Drive Losses:

Accessory drive power extraction is directly from the accessory drive pad on the reduction gearbox. Reduce SHP to prop-fan by amount of accessory power extraction at each point.

Nozzle Throat Area

The effective nozzle throat area is constant for all conditions at 300.0 in².



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Standard and Hot Day; Takeoff and Maximum Continuous

| Altitude (Ft $\times 10^{-3}$) | MACH Number | | | |
|------------------------------------|-------------|----|----|----|
| | 0 | .1 | .2 | .3 |
| 0 | X | X | X | X |

Standard Day; Maximum Climb, Maximum Continuous and Part Power to Idle

| Altitude (Ft $\times 10^{-3}$) | MACH Number | | | | | | | |
|------------------------------------|-------------|----|----|----|----|----|-----|----|
| | .2 | .3 | .4 | .5 | .6 | .7 | .75 | .8 |
| 0 | X | X | X | X | X | X | X | X |
| 5 | X | X | X | X | X | | | |
| 10 | X | X | X | X | X | X | X | X |
| 15 | | X | X | X | X | X | X | X |
| 20 | | | X | X | X | X | X | X |
| 25 | | | X | X | X | X | X | X |
| 30 | | | | X | X | X | X | X |
| 35 | | | | X | X | X | X | X |
| 40 | | | | X | X | X | X | X |
| 45 | | | | X | X | X | X | X |

Figure IV-1. Matrix of flight conditions for performance data

DETROIT DIESEL ALLISON DIVISION
 ZERO BLEED
 EDR 9774
 P0370-37 TURBOPROP
 100 PERCENT RECOVERY
 STD DAY
 C FEET ALTITUDE
 POWER MACH SHP SFC NF FA ESHP WCIN TNDZ PNDZ RC NO
 TO 1 0.0 8550. 0.443 3744. 1213.2 9404. 42.9 1638. 16.82 12.7 0001
 M.C. 0.0 7511. 0.443 3744. 1070.8 8233. 42.9 1547. 16.58 12.7 0002
 TO 1 0.0 7501. 0.443 3744. 936.7 8233. 42.9 1546. 16.59 12.7 0003
 M.C. 0.0 7501. 0.443 3744. 876.5 8233. 42.9 1546. 16.59 12.7 0004
 TO 1 0.0 7501. 0.443 3744. 876.5 8233. 42.9 1546. 16.59 12.7 0005
 M.C. 0.0 7501. 0.443 3744. 876.5 8233. 42.9 1546. 16.59 12.7 0006
 TO 1 0.0 7501. 0.443 3744. 876.5 8233. 42.9 1546. 16.59 12.7 0007
 M.C. 0.0 7501. 0.443 3744. 876.5 8233. 42.9 1546. 16.59 12.7 0008

T AMBIENT = 89.8°F

O FEET ALTITUDE

TO 1 0.0 7766. 0.443 3744. 1399.5 8595. 43.1 1573. 16.31 12.7 0009
 M.C. 0.0 7766. 0.443 3744. 1399.5 8595. 43.1 1573. 16.31 12.7 0010
 TO 1 0.0 7766. 0.443 3744. 1399.5 8595. 43.1 1573. 16.31 12.7 0011
 M.C. 0.0 7766. 0.443 3744. 1399.5 8595. 43.1 1573. 16.31 12.7 0012
 TO 1 0.0 7766. 0.443 3744. 1399.5 8595. 43.1 1573. 16.31 12.7 0013
 M.C. 0.0 7766. 0.443 3744. 1399.5 8595. 43.1 1573. 16.31 12.7 0014
 TO 1 0.0 7766. 0.443 3744. 1399.5 8595. 43.1 1573. 16.31 12.7 0015
 M.C. 0.0 7766. 0.443 3744. 1399.5 8595. 43.1 1573. 16.31 12.7 0016

GENERAL MOTORS CORPORATION

EDR 9774

PC370-37 TURBOPROP

GENERAL MOTORS CORPORATION

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

0 FEET ALTITUDE

[illegible]

T AMBIENT = 89.8°F

0 FEET ALTITUDE

[illegible]

DETROIT DIESEL ALLISON DIVISION
ZERO BLEED

12

GENERAL MOTORS CORPORATION

ZERO POWER EXTRACTION

MO

8050
8049

005000578

0063
0063
0066

0074

0-869

DETROIT DIESEL ALLISON DIVISION

ZERO BLEED

14

GENERAL MOTORS CORPORATION
ZERO POWER EXTRACTION

EDR 9774
P0370-37 TURBOPROP
100 PERCENT RECOVERY
STD DAY

03370 UNCL
DETROIT DIESEL DIVISION

[illegible]

GENERAL MOTORS CORPORATION

ZERO POWER EXTRACTION

30000 FEET ALTITUDE

**CLIMB
POWER**

ON

00173 00173 00173 00173

DETROIT DIESEL ALLISON DIVISION

EDR 9774

GENERAL MOTORS CORPORATION

PD370-37 TURBOPROP

ZERO BLEED

100 PERCENT RECOVERY

ZERO POWER EXTRACTION

STD DAY

40000 FEET ALTITUDE

| POWER | MACH | SHP | SFC | WF | FN | ESHP | WCIN | TNOZ | PNOZ | RC | NO |
|---------------|------|-----|-------|-----|-----|------|------|------|------|-----|------|
| CLIMB M.C. | 0.50 | 568 | 52900 | 309 | 025 | 075 | 44 | 11 | 375 | 098 | 0378 |
| | 0.50 | 509 | 4000 | 989 | 161 | 179 | 44 | 16 | 379 | 433 | 0379 |
| | 0.50 | 471 | 3900 | 716 | 111 | 111 | 44 | 20 | 381 | 118 | 0380 |
| | 0.50 | 409 | 3200 | 654 | 080 | 060 | 44 | 25 | 382 | 222 | 0381 |
| | 0.50 | 358 | 2200 | 508 | 050 | 052 | 44 | 30 | 383 | 333 | 0382 |
| CLIMB M.C. | 0.60 | 338 | 4000 | 562 | 191 | 089 | 44 | 03 | 384 | 333 | 0383 |
| | 0.60 | 328 | 3900 | 535 | 154 | 069 | 44 | 11 | 385 | 333 | 0384 |
| | 0.60 | 291 | 3800 | 456 | 128 | 055 | 44 | 15 | 386 | 333 | 0385 |
| | 0.60 | 261 | 3700 | 351 | 095 | 035 | 44 | 20 | 387 | 333 | 0386 |
| | 0.60 | 228 | 3600 | 255 | 065 | 025 | 44 | 25 | 388 | 333 | 0387 |
| CLIMB M.C. | 0.70 | 207 | 3500 | 169 | 035 | 015 | 44 | 30 | 389 | 333 | 0388 |
| | 0.70 | 187 | 3400 | 109 | 021 | 009 | 44 | 35 | 390 | 333 | 0389 |
| | 0.70 | 167 | 3300 | 99 | 011 | 001 | 44 | 40 | 391 | 333 | 0390 |
| CLIMB M.C. | 0.80 | 147 | 3200 | 89 | 001 | 001 | 44 | 45 | 392 | 333 | 0391 |
| | 0.80 | 127 | 3100 | 79 | 001 | 001 | 44 | 50 | 393 | 333 | 0392 |
| | 0.80 | 107 | 3000 | 69 | 001 | 001 | 44 | 55 | 394 | 333 | 0393 |
| | 0.80 | 87 | 2900 | 59 | 001 | 001 | 44 | 60 | 395 | 333 | 0394 |
| | 0.80 | 67 | 2800 | 49 | 001 | 001 | 44 | 65 | 396 | 333 | 0395 |
| CLIMB M.C. | 0.90 | 47 | 2700 | 39 | 001 | 001 | 44 | 70 | 397 | 333 | 0396 |
| | 0.90 | 27 | 2600 | 29 | 001 | 001 | 44 | 75 | 398 | 333 | 0397 |
| | 0.90 | 7 | 2500 | 19 | 001 | 001 | 44 | 80 | 399 | 333 | 0398 |
| CLIMB M.C. | 1.00 | 27 | 2400 | 9 | 001 | 001 | 44 | 85 | 400 | 333 | 0400 |
| | 1.00 | 7 | 2300 | 0 | 001 | 001 | 44 | 90 | 401 | 333 | 0401 |
| | 1.00 | 0 | 2200 | 0 | 001 | 001 | 44 | 95 | 402 | 333 | 0402 |
| | 1.00 | 0 | 2100 | 0 | 001 | 001 | 44 | 100 | 403 | 333 | 0403 |
| | 1.00 | 0 | 2000 | 0 | 001 | 001 | 44 | 105 | 404 | 333 | 0404 |
| | 1.00 | 0 | 1900 | 0 | 001 | 001 | 44 | 110 | 405 | 333 | 0405 |
| | 1.00 | 0 | 1800 | 0 | 001 | 001 | 44 | 115 | 406 | 333 | 0406 |
| | 1.00 | 0 | 1700 | 0 | 001 | 001 | 44 | 120 | 407 | 333 | 0407 |
| CLIMB M.C. | 1.10 | 0 | 1600 | 0 | 001 | 001 | 44 | 125 | 408 | 333 | 0408 |
| | 1.10 | 0 | 1500 | 0 | 001 | 001 | 44 | 130 | 409 | 333 | 0409 |
| | 1.10 | 0 | 1400 | 0 | 001 | 001 | 44 | 135 | 410 | 333 | 0410 |
| | 1.10 | 0 | 1300 | 0 | 001 | 001 | 44 | 140 | 411 | 333 | 0411 |
| | 1.10 | 0 | 1200 | 0 | 001 | 001 | 44 | 145 | 412 | 333 | 0412 |
| | 1.10 | 0 | 1100 | 0 | 001 | 001 | 44 | 150 | 413 | 333 | 0413 |
| | 1.10 | 0 | 1000 | 0 | 001 | 001 | 44 | 155 | 414 | 333 | 0414 |
| | 1.10 | 0 | 900 | 0 | 001 | 001 | 44 | 160 | 415 | 333 | 0415 |
| | 1.10 | 0 | 800 | 0 | 001 | 001 | 44 | 165 | 416 | 333 | 0416 |

DETROIT DIESEL ALLISON DIVISION

EDR 9774

GENERAL MOTORS CORPORATION

ZERO BLEED

P0370-37 TURBOPROP

ZERO POWER EXTRACTION

100 PERCENT RECOVERY

STD DAY

45000 FEET ALTITUDE

| POWER | MACH | SHF | SFC | WF | FN | ESHP | WCIN | TNOZ | PHOZ | RC | NO |
|---------------|------|-------|-----|------|------|-------|------|------|------|----|----|
| CLIMB M.C. | 0.50 | 1774. | 412 | 736. | 9.5 | 2051. | 6.6 | 511. | 65 | 0 | 17 |
| | 0.50 | 1794. | 409 | 712. | 152. | 2041. | 44. | 114. | 22 | 43 | 19 |
| | 0.50 | 1814. | 406 | 688. | 154. | 2031. | 44. | 135. | 40 | 13 | 20 |
| | 0.50 | 1834. | 403 | 664. | 156. | 2021. | 44. | 156. | 37 | 11 | 21 |
| | 0.50 | 1854. | 400 | 640. | 158. | 2011. | 44. | 177. | 34 | 9 | 22 |
| | 0.50 | 1874. | 397 | 616. | 160. | 2001. | 44. | 198. | 31 | 7 | 23 |
| | 0.50 | 1894. | 394 | 592. | 162. | 1991. | 44. | 219. | 28 | 5 | 24 |
| | 0.50 | 1914. | 391 | 568. | 164. | 1981. | 44. | 240. | 25 | 3 | 25 |
| | 0.50 | 1934. | 388 | 544. | 166. | 1971. | 44. | 261. | 22 | 1 | 26 |
| | 0.50 | 1954. | 385 | 520. | 168. | 1961. | 44. | 282. | 19 | 0 | 27 |
| | 0.50 | 1974. | 382 | 496. | 170. | 1951. | 44. | 303. | 16 | 0 | 28 |
| | 0.50 | 1994. | 379 | 472. | 172. | 1941. | 44. | 324. | 13 | 0 | 29 |
| | 0.50 | 2014. | 376 | 448. | 174. | 1931. | 44. | 345. | 10 | 0 | 30 |
| | 0.50 | 2034. | 373 | 424. | 176. | 1921. | 44. | 366. | 7 | 0 | 31 |
| | 0.50 | 2054. | 370 | 400. | 178. | 1911. | 44. | 387. | 4 | 0 | 32 |
| | 0.50 | 2074. | 367 | 376. | 180. | 1901. | 44. | 408. | 1 | 0 | 33 |
| | 0.50 | 2094. | 364 | 352. | 182. | 1891. | 44. | 429. | 0 | 0 | 34 |
| | 0.50 | 2114. | 361 | 328. | 184. | 1881. | 44. | 450. | 0 | 0 | 35 |
| | 0.50 | 2134. | 358 | 304. | 186. | 1871. | 44. | 471. | 0 | 0 | 36 |
| | 0.50 | 2154. | 355 | 280. | 188. | 1861. | 44. | 492. | 0 | 0 | 37 |
| | 0.50 | 2174. | 352 | 256. | 190. | 1851. | 44. | 513. | 0 | 0 | 38 |
| | 0.50 | 2194. | 349 | 232. | 192. | 1841. | 44. | 534. | 0 | 0 | 39 |
| | 0.50 | 2214. | 346 | 208. | 194. | 1831. | 44. | 555. | 0 | 0 | 40 |
| | 0.50 | 2234. | 343 | 184. | 196. | 1821. | 44. | 576. | 0 | 0 | 41 |
| | 0.50 | 2254. | 340 | 160. | 198. | 1811. | 44. | 597. | 0 | 0 | 42 |
| | 0.50 | 2274. | 337 | 136. | 200. | 1801. | 44. | 618. | 0 | 0 | 43 |
| | 0.50 | 2294. | 334 | 112. | 202. | 1791. | 44. | 639. | 0 | 0 | 44 |
| | 0.50 | 2314. | 331 | 88. | 204. | 1781. | 44. | 660. | 0 | 0 | 45 |
| | 0.50 | 2334. | 328 | 64. | 206. | 1771. | 44. | 681. | 0 | 0 | 46 |
| | 0.50 | 2354. | 325 | 40. | 208. | 1761. | 44. | 702. | 0 | 0 | 47 |
| | 0.50 | 2374. | 322 | 16. | 210. | 1751. | 44. | 723. | 0 | 0 | 48 |
| | 0.50 | 2394. | 319 | 0 | 212. | 1741. | 44. | 744. | 0 | 0 | 49 |
| | 0.50 | 2414. | 316 | 0 | 214. | 1731. | 44. | 765. | 0 | 0 | 50 |
| | 0.50 | 2434. | 313 | 0 | 216. | 1721. | 44. | 786. | 0 | 0 | 51 |
| | 0.50 | 2454. | 310 | 0 | 218. | 1711. | 44. | 807. | 0 | 0 | 52 |
| | 0.50 | 2474. | 307 | 0 | 220. | 1701. | 44. | 828. | 0 | 0 | 53 |
| | 0.50 | 2494. | 304 | 0 | 222. | 1691. | 44. | 849. | 0 | 0 | 54 |
| | 0.50 | 2514. | 301 | 0 | 224. | 1681. | 44. | 870. | 0 | 0 | 55 |
| | 0.50 | 2534. | 298 | 0 | 226. | 1671. | 44. | 891. | 0 | 0 | 56 |
| | 0.50 | 2554. | 295 | 0 | 228. | 1661. | 44. | 912. | 0 | 0 | 57 |
| | 0.50 | 2574. | 292 | 0 | 230. | 1651. | 44. | 933. | 0 | 0 | 58 |
| | 0.50 | 2594. | 289 | 0 | 232. | 1641. | 44. | 954. | 0 | 0 | 59 |
| | 0.50 | 2614. | 286 | 0 | 234. | 1631. | 44. | 975. | 0 | 0 | 60 |

GENERAL MOTORS CORPORATION

0 FEET ALTITUDE

22

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GENERAL MOTORS CORPORATION

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

5000 FEET ALTITUDE

POWER
CLIMB
M.C.

CLIMB
M.C.

CLIMB
M.C.

CLIMB
M.C.

CLIMB
M.C.

MACH

DEP SHD INL DEL BFM

DEL SHD PER DEL RT BLEED DEL FR

K

NO

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GENERAL MOTORS CORPORATION

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

10000 FEET ALTITUDE

[illegible]

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GENERAL MOTORS CORPORATION

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

15000 FEET ALTITUDE

NO
N
DEL SHP PER DEL AF LEO DEL FN
DEL SHP JAL EL BEF
PER SHP JAL EL BEF
MACH

POWER
CLIMB
M.C.

CLIMB
M.C.

CLIMB
M.C.

CLIMB
M.C.

CLIMB
M.C.

CLIMB
M.C.

GENERAL MOTORS CORPORATION

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SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

20000 FEET ALTITUDE

POWER
CLIMB
M.C.

CLIMB
M.C.

CLIMB
M.C.

CLIMB
M.C.

CLIMB
M.C.

MACH

REF-SPD INTL DEL TKA

DEL SPD PER DEL-HI-LEO DEL PA

K

NO

PD370-37 TURAOPROP

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

25000 FEET ALTITUDE

POWER
 INTER
 M.C.
 INTER
 M.C.
 INTER
 M.C.
 INTER
 M.C.
 INTER
 M.C.
 MACH
 DEB-REP-INT-REF
 14-13-REF
 DEL-REP-REF-AL-LED-DEL-FR
 X
 MD

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GENERAL MOTORS CORPORATION

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SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

30000 FEET ALTITUDE

| POWER | MACH | DEL SRP | INLET DEL FR | DEL SRP | PER DEL | BLEED DEL FR | K | NO |
|---------------|------|---------|--------------|---------|---------|--------------|-----|------|
| CLIMB M.C. | 0.50 | 7.2 | 3.0 | 46.1 | 1.1 | 0.8 | 1.1 | 0297 |
| | 0.50 | 5.5 | 2.8 | 45.1 | 1.0 | 0.8 | 1.0 | 0298 |
| | 0.50 | 4.2 | 2.6 | 44.1 | 0.9 | 0.8 | 0.9 | 0300 |
| | 0.50 | 3.2 | 2.4 | 43.1 | 0.8 | 0.8 | 0.8 | 0301 |
| | 0.50 | 2.5 | 2.2 | 42.1 | 0.7 | 0.8 | 0.7 | 0302 |
| | 0.50 | 2.0 | 2.0 | 41.1 | 0.6 | 0.8 | 0.6 | 0303 |
| | 0.50 | 1.6 | 1.8 | 40.1 | 0.5 | 0.8 | 0.5 | 0304 |
| | 0.50 | 1.3 | 1.6 | 39.1 | 0.4 | 0.8 | 0.4 | 0305 |
| | 0.50 | 1.0 | 1.4 | 38.1 | 0.3 | 0.8 | 0.3 | 0306 |
| | 0.50 | 0.8 | 1.2 | 37.1 | 0.2 | 0.8 | 0.2 | 0307 |
| | 0.50 | 0.6 | 1.0 | 36.1 | 0.1 | 0.8 | 0.1 | 0308 |
| | 0.50 | 0.5 | 0.9 | 35.1 | 0.1 | 0.8 | 0.1 | 0309 |
| | 0.50 | 0.4 | 0.8 | 34.1 | 0.1 | 0.8 | 0.1 | 0310 |
| | 0.50 | 0.3 | 0.7 | 33.1 | 0.1 | 0.8 | 0.1 | 0311 |
| | 0.50 | 0.2 | 0.6 | 32.1 | 0.1 | 0.8 | 0.1 | 0312 |
| | 0.50 | 0.2 | 0.5 | 31.1 | 0.1 | 0.8 | 0.1 | 0313 |
| | 0.50 | 0.1 | 0.4 | 30.1 | 0.1 | 0.8 | 0.1 | 0314 |
| | 0.50 | 0.1 | 0.3 | 29.1 | 0.1 | 0.8 | 0.1 | 0315 |
| | 0.50 | 0.1 | 0.2 | 28.1 | 0.1 | 0.8 | 0.1 | 0316 |
| | 0.50 | 0.1 | 0.1 | 27.1 | 0.1 | 0.8 | 0.1 | 0317 |
| | 0.50 | 0.1 | 0.1 | 26.1 | 0.1 | 0.8 | 0.1 | 0318 |
| | 0.50 | 0.1 | 0.1 | 25.1 | 0.1 | 0.8 | 0.1 | 0319 |
| | 0.50 | 0.1 | 0.1 | 24.1 | 0.1 | 0.8 | 0.1 | 0320 |
| | 0.50 | 0.1 | 0.1 | 23.1 | 0.1 | 0.8 | 0.1 | 0321 |
| | 0.50 | 0.1 | 0.1 | 22.1 | 0.1 | 0.8 | 0.1 | 0322 |
| | 0.50 | 0.1 | 0.1 | 21.1 | 0.1 | 0.8 | 0.1 | 0323 |
| | 0.50 | 0.1 | 0.1 | 20.1 | 0.1 | 0.8 | 0.1 | 0324 |
| | 0.50 | 0.1 | 0.1 | 19.1 | 0.1 | 0.8 | 0.1 | 0325 |
| | 0.50 | 0.1 | 0.1 | 18.1 | 0.1 | 0.8 | 0.1 | 0326 |
| | 0.50 | 0.1 | 0.1 | 17.1 | 0.1 | 0.8 | 0.1 | 0327 |
| | 0.50 | 0.1 | 0.1 | 16.1 | 0.1 | 0.8 | 0.1 | 0328 |
| | 0.50 | 0.1 | 0.1 | 15.1 | 0.1 | 0.8 | 0.1 | 0329 |
| | 0.50 | 0.1 | 0.1 | 14.1 | 0.1 | 0.8 | 0.1 | 0330 |
| | 0.50 | 0.1 | 0.1 | 13.1 | 0.1 | 0.8 | 0.1 | 0331 |
| | 0.50 | 0.1 | 0.1 | 12.1 | 0.1 | 0.8 | 0.1 | 0332 |
| | 0.50 | 0.1 | 0.1 | 11.1 | 0.1 | 0.8 | 0.1 | 0333 |
| | 0.50 | 0.1 | 0.1 | 10.1 | 0.1 | 0.8 | 0.1 | 0334 |
| | 0.50 | 0.1 | 0.1 | 9.1 | 0.1 | 0.8 | 0.1 | 0335 |
| | 0.50 | 0.1 | 0.1 | 8.1 | 0.1 | 0.8 | 0.1 | 0336 |
| | 0.50 | 0.1 | 0.1 | 7.1 | 0.1 | 0.8 | 0.1 | 0337 |
| | 0.50 | 0.1 | 0.1 | 6.1 | 0.1 | 0.8 | 0.1 | 0338 |
| | 0.50 | 0.1 | 0.1 | 5.1 | 0.1 | 0.8 | 0.1 | 0339 |
| | 0.50 | 0.1 | 0.1 | 4.1 | 0.1 | 0.8 | 0.1 | 0340 |
| | 0.50 | 0.1 | 0.1 | 3.1 | 0.1 | 0.8 | 0.1 | 0341 |
| | 0.50 | 0.1 | 0.1 | 2.1 | 0.1 | 0.8 | 0.1 | 0342 |
| | 0.50 | 0.1 | 0.1 | 1.1 | 0.1 | 0.8 | 0.1 | 0343 |
| | 0.50 | 0.1 | 0.1 | 0.1 | 0.1 | 0.8 | 0.1 | 0344 |

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GENERAL MOTORS CORPORATION

P0370-37 TURBOPROP

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

35000 FEET ALTITUDE

| POWER CLIMB M.C. | MACH | PER SHP DEL SHP | INLET DEL IN | PER SHP DEL SHP | BLEED DEL BLEED | K | NO |
|------------------------|------|--------------------|-----------------|--------------------|--------------------|------|------|
| 0.50 | 0.50 | 37.2 | 6.1 | 37.2 | 3.1 | 0.1 | 0337 |
| 0.50 | 0.50 | 36.6 | 5.7 | 36.6 | 3.0 | 0.2 | 0338 |
| 0.50 | 0.50 | 35.0 | 5.3 | 35.0 | 2.9 | 0.3 | 0339 |
| 0.50 | 0.50 | 33.5 | 4.9 | 33.5 | 2.8 | 0.4 | 0340 |
| 0.50 | 0.50 | 32.0 | 4.5 | 32.0 | 2.7 | 0.5 | 0341 |
| 0.50 | 0.50 | 30.5 | 4.1 | 30.5 | 2.6 | 0.6 | 0342 |
| 0.50 | 0.50 | 29.0 | 3.7 | 29.0 | 2.5 | 0.7 | 0343 |
| 0.50 | 0.50 | 27.5 | 3.3 | 27.5 | 2.4 | 0.8 | 0344 |
| 0.50 | 0.50 | 26.0 | 2.9 | 26.0 | 2.3 | 0.9 | 0345 |
| 0.50 | 0.50 | 24.5 | 2.5 | 24.5 | 2.2 | 1.0 | 0346 |
| 0.50 | 0.50 | 23.0 | 2.1 | 23.0 | 2.1 | 1.1 | 0347 |
| 0.50 | 0.50 | 21.5 | 1.7 | 21.5 | 2.0 | 1.2 | 0348 |
| 0.50 | 0.50 | 20.0 | 1.3 | 20.0 | 1.9 | 1.3 | 0349 |
| 0.50 | 0.50 | 18.5 | 0.9 | 18.5 | 1.8 | 1.4 | 0350 |
| 0.50 | 0.50 | 17.0 | 0.5 | 17.0 | 1.7 | 1.5 | 0351 |
| 0.50 | 0.50 | 15.5 | 0.1 | 15.5 | 1.6 | 1.6 | 0352 |
| 0.50 | 0.50 | 14.0 | 0.0 | 14.0 | 1.5 | 1.7 | 0353 |
| 0.50 | 0.50 | 12.5 | 0.0 | 12.5 | 1.4 | 1.8 | 0354 |
| 0.50 | 0.50 | 11.0 | 0.0 | 11.0 | 1.3 | 1.9 | 0355 |
| 0.50 | 0.50 | 9.5 | 0.0 | 9.5 | 1.2 | 2.0 | 0356 |
| 0.50 | 0.50 | 8.0 | 0.0 | 8.0 | 1.1 | 2.1 | 0357 |
| 0.50 | 0.50 | 6.5 | 0.0 | 6.5 | 1.0 | 2.2 | 0358 |
| 0.50 | 0.50 | 5.0 | 0.0 | 5.0 | 0.9 | 2.3 | 0359 |
| 0.50 | 0.50 | 3.5 | 0.0 | 3.5 | 0.8 | 2.4 | 0360 |
| 0.50 | 0.50 | 2.0 | 0.0 | 2.0 | 0.7 | 2.5 | 0361 |
| 0.50 | 0.50 | 0.5 | 0.0 | 0.5 | 0.6 | 2.6 | 0362 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.5 | 2.7 | 0363 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.4 | 2.8 | 0364 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.3 | 2.9 | 0365 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.2 | 3.0 | 0366 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.1 | 3.1 | 0367 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 3.2 | 0368 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 3.3 | 0369 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 3.4 | 0370 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 3.5 | 0371 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 3.6 | 0372 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 3.7 | 0373 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 3.8 | 0374 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 3.9 | 0375 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 4.0 | 0376 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 4.1 | 0377 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 4.2 | 0378 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 4.3 | 0379 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 4.4 | 0380 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 4.5 | 0381 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 4.6 | 0382 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 4.7 | 0383 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 4.8 | 0384 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 4.9 | 0385 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 5.0 | 0386 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 5.1 | 0387 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 5.2 | 0388 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 5.3 | 0389 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 5.4 | 0390 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 5.5 | 0391 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 5.6 | 0392 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 5.7 | 0393 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 5.8 | 0394 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 5.9 | 0395 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 6.0 | 0396 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 6.1 | 0397 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 6.2 | 0398 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 6.3 | 0399 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 6.4 | 0400 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 6.5 | 0401 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 6.6 | 0402 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 6.7 | 0403 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 6.8 | 0404 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 6.9 | 0405 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 7.0 | 0406 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 7.1 | 0407 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 7.2 | 0408 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 7.3 | 0409 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 7.4 | 0410 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 7.5 | 0411 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 7.6 | 0412 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 7.7 | 0413 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 7.8 | 0414 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 7.9 | 0415 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 8.0 | 0416 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 8.1 | 0417 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 8.2 | 0418 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 8.3 | 0419 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 8.4 | 0420 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 8.5 | 0421 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 8.6 | 0422 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 8.7 | 0423 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 8.8 | 0424 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 8.9 | 0425 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 9.0 | 0426 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 9.1 | 0427 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 9.2 | 0428 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 9.3 | 0429 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 9.4 | 0430 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 9.5 | 0431 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 9.6 | 0432 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 9.7 | 0433 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 9.8 | 0434 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 9.9 | 0435 |
| 0.50 | 0.50 | 0.0 | 0.0 | 0.0 | 0.0 | 10.0 | 0436 |

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SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS
45000 FEET ALTITUDE

Diagram illustrating a 1000-bit data stream divided into 100 segments of 10 bits each. The segments are labeled on the left as POWER, CLIMB, CLIMB, CLIMB, CLIMB, CLIMB, CLIMB, CLIMB, CLIMB, and CLIMB. The segments are also labeled on the right as MACH, DEL-STA, DEL-STA, DEL-STA, DEL-STA, DEL-STA, DEL-STA, DEL-STA, DEL-STA, and DEL-STA. The diagram shows a sequence of bits (0s and 1s) forming a data stream.